# Hygrothermal effects on the behavior of reinforced-concrete beams strengthened by bonded composite laminate plates

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**Abstract.** The purpose of this paper is to investigate the hygrothermal effects on the behavior of reinforced-concrete beams strengthened by bonded composite laminate plates ( $\theta$ n/90 m)s. This work is based on a simple theoretical model to estimate the interfacial stresses developed between the concrete beam and the composite with taking into account the hygrothermal effect. Fibre orientation angle effects of number of 90° layers and effects of plate thickness and length on the distributions of interfacial stress in the concrete beams reinforced with composite plates have also been studied.

Keywords: interfacial stresses; concrete beam; composite laminate; hygrothermal effects; fibre angle

# 1. Introduction

The plate bonding technique is used to increase the strength and the stiffness or repair the existing reinforced concrete structure. The use of the composite fiberreinforced plastic (FRP) become more and more very effective given its simplicity. Many studies have been conducted, to predict the interfacial stresses, see, for example, those by Tounsi et al. (2007), Tounsi (2006), Benyoucef et al. (2006), Vilnay (1988), Roberts (1989), Roberts et al. (1989), Malek et al. (1994), Robinovitch et al. (2000), Ye (2001), Smith et al. (2001), Barnes et al. (2001), Stratford et al. (2006). Bouazaoui (2008) have studied the interfacial shear strength between the steel bar surface and concrete surface of steel rods bonded into concrete. Many approximate closed-form solutions have been developed in the past decade for the interfacial stresses in beams bonded with a steel or FRP plate (Vilnay 1988, Roberts 1989, Roberts et al. 1989, Taljsten 1997, Smith et al. 2001, Tounsi 2006).

The solution presented by Smith *et al.* (2001) seems to be the more accurate widely applicable solution, particularly when the flexural stiffness of the bonded plate becomes significant. Rabinovich *et al.* (2000) has presented a higher order analysis in which the adhesive layer was treated as an elastic medium with negligible longitudinal stiffness. This leads to uniform stresses and linearly varying normal stresses through the thickness of the adhesive layer. The significance of their solution is that it is the first solution that satisfies the stress-free boundary condition at the ends of the adhesive layer. Using the same approach, they investigated the effects of an uneven adhesive layer Rabinovich et al. (2001). Shen et al. (2001) proposed an alternative analytical complementary energy approach, which resulted in closed-form expressions. Recently, many authors have conducted a numerical study in different directions to illustrate the principal parameters in order to estimate the distributions of interfacial stress in beams reinforced with composite plates (Daouadji 2017, Bouakaz et al. 2014, Krour et al. 2014, Touati et al. 2015, Hadji et al. 2016, Kara 2016, Elamary et al. 2016). The analytical models present often the assumption of constant environment conditions, while the RC beam and the FRP are subjected to changing temperature and moisture conditions and it should be including in the analysis Gibson (1994). In this paper, the hygrothermal effects in the concrete beam and the soffit plate will be included to estimate the interfacial shear and normal stresses. For this case, we introduce an analytical solution which include the mechanical properties of the beam, the plate and the adhesive layer under thermal (temperature effect) and hygroscopic (moisture effect) conditions. The most used solution is Teng's solution Smith et al. (2001). This solution doesn't consider the fibre orientation contrary to Tounsi (2006) solution which will be used in this paper. The total interfacial stress is the sum of Tounsi solution and the additional one due to the hygrothermal deformation of the beam and the plate (Bouderba et al. 2013, Bousahla et al. 2016, Beldjelili et al. 2016, Menasria et al. 2017, Chikh et al. 2017).

Rabinovich et al. (2001) and material nonlinearity

We can use more advanced theory, for example, those by El-Haina *et al.* (2017), Younsi *et al.* (2018), Bouhadra *et al.* (2018), Benchohra *et al.* (2018), Bourada *et al.* (2018).

Results have been presented to show the hygro-thermal effect on the interfacial shear and normal stresses.

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# 2. Theoretical modelling





Fig. 2 Differential segment of a soffit-plated beam

As shown in Fig. 1, a concrete beam (Adherend 1) strengthened by FRP plate (Adherend 2) and bounded by an adhesive layer is considered. This beam is simply supported reinforced beam and subjected to a uniform distributed load. Geometry and cross-sections are shown in Fig. 1.

The following assumptions are used:

• The materials concrete beam, FRP plate and adhesive are linear elastic.

• Shear and normal stresses in the adhesive layer are constant across its thickness.

• The curvature in the beam and the plate are same.

## 2.1 Governing differential equations

The Fig. 2 represents a differential segment of plated beam.  $\tau(x)$  and  $\sigma(x)$  are the interfacial shear and normal stresses respectively with positive sign convention for the bending moment, shear force, axial force and applied loading. The derivation of the new solution below is described in terms of adherends 1 and 2, where adherend 1 is the beam and adherend 2 is the soffit plate.

The shear strain  $\gamma$  in the adhesive layer can be written as

$$\gamma = \frac{du(x, y)}{dy} + \frac{dv(x, y)}{dx} \tag{1}$$

u(x,y) and v(x,y) are the horizontal and vertical displacements of the adhesive layer respectively.  $\tau(x)$  is given as

$$\tau(x) = G_a \left( \frac{du(x, y)}{dy} + \frac{dv(x, y)}{dx} \right)$$
(2)

where  $G_a$  is the shear modulus of the adhesive layer. Differentiating the expression (2) with respect to x gives

$$\frac{d\tau(x)}{dx} = G_a \left( \frac{d^2 u(x, y)}{dy} + \frac{d^2 v(x, y)}{dx} \right)$$
(3)

The curvature is function of the applied moment  $M_T(x)$ 

$$\frac{d^2 v(x)}{dx^2} = -\frac{1}{\left(EI\right)_t} M_T(x) \tag{4}$$

where  $(EI)_t$  is the total flexural rigidity of the composite section. u(x,y) must vary linearly across the adhesive thickness  $t_a$ , then

$$\frac{du}{dy} = \frac{1}{t_a} \left[ u_2(x) - u_1(x) \right]$$
(5)

and

$$\frac{d^2u(x,y)}{dxdy} = \frac{1}{t_a} \left( \frac{du_2(x)}{dx} - \frac{du_1(x)}{dx} \right)$$
(6)

where  $u_1(x)$  and  $u_2(x)$  are the longitudinal displacements at the base of adherend 1 and the top of adherend 2, respectively. Eq. (3) can be rewritten as

$$\frac{d\tau(x)}{dx} = G_a \left( \frac{du_2(x)}{dx} - \frac{du_1(x)}{dx} - \frac{t_a}{(EI)_t} M_T(x) \right)$$
(7)

The third term in parentheses in Eq. (7) can be ignored Smith *et al.* (2001) in the following derivation. The strains at the base of adherend 1 and the top of adherend 2 taking account the hygrothermal deformations are given as

$$\varepsilon_{1}(x) = \frac{du_{1}(x)}{dx} = \frac{y_{1}}{E_{1}I_{1}}M_{1}(x) - \frac{1}{E_{1}A_{1}}N_{1}(x) + \alpha_{1}\Delta T + \beta_{1}\Delta C$$
(8)

Where  $\alpha_1$  is the coefficient of thermal expansion and  $\beta_1$  is the coefficient of hygroscopic expansion of the RC beam.  $\Delta T$  and  $\Delta C$  are the temperature and percent moisture change respectively.

The laminate theory is used to estimate the strain of the symmetrical composite plate (Herakovich 1998), i.e.,

$$\varepsilon_x^0 = A_{11}N_x \frac{1}{b_2}$$
 and  $k_x = D_{11}M_x \frac{1}{b_2}$  (9)

[A'] is the inverse of the extensional matrix [A]; [D'] is the inverse of the flexural matrix [D];  $b_2$  is a width of FRP plate. Using CLT, the strain at the top of the FRP plate 2 is given as

$$\varepsilon_2(x) = \varepsilon_x^0 - k_x \frac{t_2}{2} + \alpha_2 \Delta T + \beta_2 \Delta C \tag{10}$$

Substituting Eqs. (9) in (10) gives the following equation

$$\varepsilon_2(x) = -D_{11} \frac{t_2}{2b_2} M_2(x) + \frac{A_{11}}{b_2} N_2(x) + \alpha_2 \Delta T + \beta_2 \Delta C \quad (11)$$

 $\alpha_2$  is the coefficient of thermal expansion and  $\beta_2$  is the coefficient of hygroscopic expansion of the plate soffit.

Considering horizontal equilibrium gives

$$\frac{dN_1(x)}{dx} = \frac{dN_2(x)}{dx} = b_2 \tau(x) \tag{12}$$

where

$$N_1(x) = N_2(x) = N(x) = b_2 \int_0^x \tau(x) dx$$
(13)

The moments in the two adherends can be related as

$$M_1(x) = RM_2(x) \tag{14}$$

with

$$R = \frac{E_1 I_1}{E_2 I_2} \tag{15}$$

and the moment equilibrium of the differential segment of the plated beam in Fig. 2 gives

$$M_T(x) = M_1(x) + M_2(x) + N(x)(y_1 + y_2 + t_a)$$
(16)

The bending moment in each adherend is function of the total applied moment and the interfacial shear stress as

$$M_{1}(x) = \frac{R}{R+1} \left[ M_{T}(x) - b_{2} \int_{0}^{x} \tau(x)(y_{1} + y_{2} + t_{a}) dx \right]$$
(17)

$$M_{2}(x) = \frac{1}{R+1} \left[ M_{T}(x) - b_{2} \int_{0}^{x} \tau(x)(y_{1} + y_{2} + t_{a}) dx \right]$$
(18)

The first derivative of the bending moment in each adherend gives

$$\frac{dM_1(x)}{dx} = V_1(x) = \frac{R}{R+1} \left[ V_T(x) - b_2 \tau(x)(y_1 + y_2 + t_a) \right]$$
(19)

$$\frac{dM_2(x)}{dx} = V_2(x) = \frac{1}{R+1} \left[ V_T(x) - b_2 \tau(x) (y_1 + y_2 + t_a) \right]$$
(20)

Substituting Eqs. (8) and (11) into Eq. (7) and differentiating the resulting equation once yields

$$\frac{d^{2}\tau(x)}{dx^{2}} = \frac{G_{a}}{t_{a}} \left( \frac{-t_{2}}{2b_{2}} D_{11}^{'} \frac{dM_{2}(x)}{dx} + \frac{A_{11}^{'}}{b_{2}} \frac{dN_{2}(x)}{dx} - \frac{y_{1}}{E_{1}I_{1}} \frac{dM_{1}(x)}{dx} + \frac{1}{E_{1}A_{1}} \frac{dN_{1}(x)}{dx} \right)$$
(21)

Substitution of the shear forces (Eqs. (19) and (20)) and axial forces (Eq. (13)) in both adherends into Eq. (21) gives the following governing differential equation for the interfacial shear stress

$$\frac{d^{2}\tau(\mathbf{x})}{d\mathbf{x}^{2}} - \frac{G_{a}}{t_{a}} \left( A_{11}^{'} + \frac{b_{2}}{E_{1}I_{1}} + \frac{(y_{1} + y_{2})(y_{1} + y_{2} + t_{a})}{E_{1}I_{1}D_{11}^{'} + b_{2}} b_{2}D_{11}^{'}\tau(\mathbf{x}) \right) + \frac{G_{a}}{t_{a}} \left( \frac{y_{1} + y_{2}}{E_{1}I_{1}D_{11}^{'} + b_{2}} D_{11}^{'} \right) V_{T}(\mathbf{x}) = 0$$
(22)

For simplicity and for such loading,  $d_2V_T(x)/dx_2=0$ , the

general solution of Eq. (22) is given by

$$\tau(x) = B_1 \cosh(\lambda x) + B_2 \sinh(\lambda x) + m_1 V_T(x)$$
(23)

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where

$$\lambda^{2} = \frac{G_{a}}{t_{a}} \left( A_{11}^{'} + \frac{b_{2}}{E_{1}A_{1}} + \frac{(y_{1} + y_{2})(y_{1} + y_{2} + t_{a})}{E_{1}I_{1}D_{11}^{'} + b_{2}} b_{2}D_{11}^{'} \right)$$
(24)

and

$$m_{1} = \frac{G_{a}}{t_{a}\lambda^{2}} \left( \frac{y_{1} + y_{2}}{E_{1}I_{1}D_{11}^{'} + b_{2}} D_{11}^{'} \right)$$
(25)

The general solution for the interfacial shear stress for a simply supported beam subjected to a uniformly distributed load is given as

$$\tau(x) = B_1 \cosh(\lambda x) + B_2 \sinh(\lambda x) + m_1 q \left(\frac{L}{2} - x - a\right)$$
(26)

The constants of integration need to be determined by applying suitable boundary conditions.

At x=0, here the moment at the plate end  $M_2(0)=N_1(0)=N_2(0)=0$  and as a result

$$M_1(0) = M_T(0) = \frac{qa}{2} (L - a)$$
(27)

Substituting Eqs. (8) and (11) into Eq. (7) with the third term ignored, and applying the above boundary condition, gives

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$$\left. \frac{d\tau(x)}{dx} \right|_{x=0} = -m_2 M_T(0) \tag{28}$$

where

$$m_2 = \frac{G_a y_1}{t_a E_1 I_1}$$
(29)

By substituting Eq. (23) into Eq. (28), B2 can be determined as

$$B_{2} = -\frac{m_{2}qa}{2\lambda}(L-a) + \frac{m_{1}}{\lambda}q + \frac{G_{a}}{t_{a}\lambda}[(\alpha_{1}-\alpha_{2})\Delta T + (\beta_{1}-\beta_{2})\Delta C] = -B_{1}$$
(30)

The normal stress in the adhesive layer,  $\sigma(x)$ , is given as

$$\sigma(x) = \frac{E_a}{t_a} \left[ v_2(x) - v_1(x) \right] \tag{31}$$

where  $v_1(x)$  and  $v_2(x)$  are the vertical displacements of adherend 1 and 2, respectively.

Differentiating Eq. (31) twice results in

$$\frac{d^2\sigma(x)}{dx^2} = K_n \left[ \frac{d^2v_1(x)}{dx^2} - \frac{d^2v_2(x)}{dx^2} \right]$$
(32)

Considering the moment-curvature relationships for the beam to be strengthened and the external reinforcement, respectively

$$\frac{d^2 v_1(x)}{dx^2} = \frac{M_1(x)}{E_1 I_1} \frac{d^2 v_2(x)}{dx^2} = -\frac{D_{11} M_2(x)}{dx^2}$$
(33)

The equilibrium of adherend 1 and 2, leads to the following relationships:

Adherend 1

$$\frac{dM_1(x)}{dx} = V_1(x) - b_2 y_1 \tau(x)$$
and
$$\frac{dV_1(x)}{dx} = -b_2 y_1 \sigma(x) - q$$
(34)

Adherend 2

$$\frac{dM_2(x)}{dx} = V_2(x) - b_2 y_2 \tau(x)$$
and
$$\frac{dV_2(x)}{dx} = -b_2 \sigma(x)$$
(35)

Based on the above equilibrium equations, the governing differential equations for the deflection of adherends 1 and 2, expressed in terms of the interfacial shear and normal stresses, are given as follows:

Adherend 1

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$$\frac{d^4 v_1(x)}{dx^4} = -\frac{1}{E_1 I_1} b_2 \sigma(x) + \frac{y_1}{E_1 I_1} b_2 \frac{d\tau(x)}{dx} + \frac{q}{E_1 I_1}$$
(36)

Adherend 2

$$\frac{d^4 v_2(x)}{dx^4} = -D_{11}' \sigma(x) + D_{11}' y_2 \frac{d\tau(x)}{dx} + \frac{q}{E_1 I_1}$$
(37)

Substitution of Eqs. (36) and (37) into the fourth derivation of the interfacial normal stress obtainable from Eq. (31) gives the following governing differential equation for the interfacial normal stress

$$\frac{d^{4}\sigma(x)}{dx^{4}} + \frac{E_{a}}{t_{a}} \left( D_{11}^{'} + \frac{b_{2}}{E_{1}I_{1}} \right) \sigma(x) - \frac{E_{a}}{t_{a}} \left( D_{11}^{'}y_{2} - \frac{y_{1}b_{2}}{E_{1}I_{1}} \right) \frac{d\tau(x)}{dx} + \frac{qE_{a}}{t_{a}E_{1}I_{1}}$$
(38)

The general solution to this fourth-order differential equation is

$$\sigma(x) = e^{-\beta x} [C_1 \cos(\beta x) + C_2 \sin(\beta x)] + e^{-\beta x} [C_3 \cos(\beta x) + C_4 \sin(\beta x)] - n_1 \frac{d\tau(x)}{dx} - n_2 q$$
(39)

For large values of x it is assumed that the normal stress approaches zero, and as a result  $C_3=C_4=0$ . The general

solution therefore becomes

$$\sigma(x) = e^{-\beta x} \left[ C_1 \cos(\beta x) + C_2 \sin(\beta x) \right]$$
  
+  $n_1 \frac{d\tau(x)}{dx} - n_2 q$  (40)

where

$$\beta = \sqrt[4]{\frac{E_a}{4t_a} \left(\frac{b_2}{E_1 I_1} + D_{11}^{'}\right)}$$
(41)

and

$$n_1 = \frac{y_1 b_2 - D_{11} E_1 I_1 y_1}{D_{11} E_1 I_1 + b_2}$$
(42)

$$a_2 = \frac{1}{D_{11}E_1I_1 + b_2} \tag{43}$$

As is described by Smith *et al.* (2001), the constants  $C_1$  and  $C_2$  in Eq. (40) are determined using the appropriate boundary conditions

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$$C_{1} = \frac{E_{a}}{2\beta^{2}t_{a}} \frac{1}{E_{1}I_{1}} \left[ V_{T}(0) + \beta M_{T}(0) \right] - \frac{n_{3}}{2\beta^{3}} \tau(0) + \frac{n_{1}}{2\beta^{3}} \left( \frac{d^{4}\tau(x)}{dx^{4}} \right|_{x=0} + \beta \frac{d^{3}\tau(x)}{dx^{3}} \right|_{x=0}$$
(44)

$$C_{2} = \frac{E_{a}}{2\beta^{2}t_{a}} \frac{1}{E_{1}I_{1}} M_{T}(0) - \frac{n_{1}}{2\beta^{2}} \frac{d^{3}\tau(x)}{dx^{3}} \bigg|_{x=0}$$
(45)

where

$$n_{3} = \frac{E_{a}b_{2}}{t_{a}} \left( \frac{y_{1}}{E_{1}I_{1}} - \frac{D_{11}t_{2}}{2b_{2}} \right)$$
(46)

# 3. Results and discussion

Numerical results are presented to illustrate and examine both thermal and hygroscopic effects on the interfacial shear and normal stresses. We consider for this work an RC beam of 3000 mm of length, a soffit plate of  $L_P=2400$  mm, a uniform distributed load q=50 KN/ml. The other geometric parameters and mechanicals properties are resumed in Table 1. All results are given for two cases of temperature and moisture change:

- ΔT=0°C, ΔC=0%
- ΔT=50°C, ΔC=1%
- The hygrothermal properties used in this study are:
- $\alpha_1 = 11*10^{-6}$  C and  $\beta_1 = *10^{-4}$  (Edward G.N. (2008)),

•  $\alpha_2=9*10^{-6}$ /°C (Schmit 1998) and  $\beta_2=1.89*10^{-3}$  (Vaddadi *et al.* 2007)

Figs. 3 to 12 shows the hygrothermal effect for various parameters. The maximum interfacial stresses increase

Component	Width (mm)	Depth (mm)	Young's Modulus (GPa)	Poisson's ratio
RC beam	200	300	30	
Soffit plate	200	4	50	
Adhesive layer	200	4	3	0.35

Table 1 Geometric and mechanicals properties

Table 2 Maximum interfacial shear and normal stresses

	$ au_{\max}(MPa)$	$\sigma_{\rm max}({ m MPa})$
Denton et al. (2001)	2.418	1.982
Deng et al. (2004)	2.410	2.007
Present (0/90)s	2.959	1.392



Fig. 3 Effect of fibre orientations on shear stress for an RC beam with a bonded composites laminates plate  $[\theta/90]_s$  for different hygroscopic cases

considerably with considering the hygrothermal effect.

#### 3.1 Comparison studies

In order to validate the accuracy of the present method, a comparison has been carried out with previously published results by Denton *et al.* (2001) and Deng *et al.* (2004) for  $\Delta T$ =50°C. Comparison results of the maximum interfacial shear and normal stresses are shown in Tables 2. It can be observed a good agreement between the present study and those given by Denton *et al.* (2001) and Deng *et al.* (2004).

## 3.2 Fibre orientation

The fibre orientation affects significantly the development of interfacial stresses. It's shown in Figs. 3 and 4 that the shear and normal stresses decrease when the angle of orientation increases.

## 3.3 Plate thickness

Figs. 5 and 6 show the effects of the plate thickness on the interfacial stresses. It seen that this plate affects considerably the normal stress and hardly the shear stress concentration. The normal stress increase with increasing the plate thickness.



Fig. 4 Effect of fibre orientations on normal stress for an RC beam with a bonded composites laminates plate  $[\theta/90]_s$  for different hygroscopic cases



Fig. 5 Effect of plate thickness on shear stress for an RC beam with a bonded composites laminates plate  $[0/90]_s$  under hygrothermal effect



Fig. 6 Effect of plate thickness on normal stress for an RC beam with a bonded composites laminates plate  $[0/90]_s$  under hygrothermal effect

## 3.4 Adhesive layer thickness

It's shown in Figs. 7 and 8 that the level of interfacial stresses is influenced considerably by the thickness of the adhesive layer. The interface shear and normal stresses decrease with increasing the thickness of adhesive layer.



Fig. 7 Effect of adhesive layer thickness on shear stress for an RC beam with a bonded composites laminates plate  $[0/90]_s$  under hygrothermal effect



Fig. 8 Effect of adhesive layer thickness on normal stress for an RC beam with a bonded composites laminates plate  $[0/90]_s$  under hygrothermal effect



Fig. 9 Effect of external layer number on shear stress for an RC beam with a bonded composites laminates plate  $[0/90_m]_s$  under hygrothermal effect

## 3.5 Number of laminate layers

The number of laminate layers have an important effect in the distribution of the interfacial stresses. Figs. 9 and 10 show a decreasing in both shear and normal stresses with increasing the number of external layers. On the other



Fig. 10 Effect of external layer number on normal stress for an RC beam with a bonded composites laminates plate  $[0/90_m]_s$  under hygrothermal effect



Fig. 11 Effect of internal layer number on shear stress for an RC beam with a bonded composites laminates plate  $[0_n/90]_s$  under hygrothermal effect



Fig. 12 Effect of internal layer number on normal stress for an RC beam with a bonded composites laminates plate  $[0_n/90]_s$  under hygrothermal effect

hand, it's shown in Figs. 11 and 12 that normal and shear stresses increase when the number of internal layers increase.

# 4. Conclusions

A simple solution has been presented in this paper in

order to calculate the interfacial stresses of retrofitted RC beam strengthened with a soffit plate under hygrothermal and mechanical loads with taking into account the fibre orientation. The following are the main conclusions.

• The inclusion of hygrothermal effect increases considerably the maximum interfacial shear and normal stresses.

• It's found that increasing the angle of fibre orientation reduces the interfacial stresses.

• The interfacial stresses increase with increasing the thickness of the FRP plate. No change has been found in the distribution of the shear stresses.

• Increasing the thickness of adhesive layer has a great role in the reduction of the interfacial shear and normal stresses.

• The interfacial stresses decrease with increasing the number of external layers and increases with increasing the number of internal layers.

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