

## Bending analysis of functionally graded thick plates with in-plane stiffness variation

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**Abstract.** In the present paper, functionally graded (FG) materials are presented to investigate the bending analysis of simply supported plates. It is assumed that the material properties of the plate vary through their length according to the power-law form. The displacement field of the present model is selected based on quasi-3D hyperbolic shear deformation theory. By splitting the deflection into bending, shear and stretching parts, the number of unknowns and equations of motion of the present formulation is reduced and hence makes them simple to use. Governing equations are derived from the principle of virtual displacements. Numerical results for deflections and stresses of powerly graded plates under simply supported boundary conditions are presented. The accuracy of the present formulation is demonstrated by comparing the computed results with those available in the literature. As conclusion, this theory is as accurate as other shear deformation theories and so it becomes more attractive due to smaller number of unknowns. Some numerical results are provided to examine the effects of the material gradation, shear deformation on the static behavior of FG plates with variation of material stiffness through their length.

**Keywords:** FGM; bending; thick plate

### 1. Introduction

The concept of functionally graded materials (FGMs) was first introduced by a group of Japanese scientists in 1984 (Yamanouchi 1990, Koizumi 1993). FGMs are a type of advanced composite materials whose properties vary gradually and continuously from one surface to another. Because of this advantage, they have been regarded as one of the advanced inhomogeneous composite materials in many engineering sectors. FGMs are made from a mixture of ceramic and metal. Ceramic constituent provides the high temperature resistance due to its low thermal conductivity and metal constituent resists the failure of the structure. The FGM is widely used in many structural applications such as aerospace, nuclear, civil, nanostructures, and automotive (Janghorban and Zare 2011, Nami and Janghorban 2014, Mouffoki *et al.* 2017, Karami *et al.* 2018a, b, c, Dash *et al.* 2018). The benefit of advanced materials and structures can be demonstrated in other applications such as (Mehtar *et al.*

2018, Henderson *et al.* 2018, Hirwani and Panda 2018, Bouadi *et al.* 2018, Cherif *et al.* 2018, Kadari *et al.* 2018, Belabed *et al.* 2018, Hajmohammad *et al.* 2018a, Mehar and Panda 2018, Karami *et al.* 2018d, e, f, g, Yazid *et al.* 2018, Youcef *et al.* 2018, Karami *et al.* 2018h, i, Shahsavari *et al.* 2017, Mehar *et al.* 2017a, b, c, Mahapatra *et al.* 2017a, b, Katariya *et al.* 2017a, Dutta *et al.* 2017, Sahoo *et al.* 2017, Beldjelili *et al.* 2016, Arani and Kolahchi 2016, Ahouel *et al.* 2016, Zemri *et al.* 2015, Bourada *et al.* 2015).

The increase in FGM application requires more accurate plate theories to predict their responses. The classical plate theory (CPT) shows inaccurate results for thick and moderately thick plates as it does not consider shear deformation. The First-order shear deformation theories (FPT) (Mindlin 1951, Reissner 1945) consider the transverse shear deformation effects. However, a shear correction factor is needed to satisfy the zero transverse shear stress boundary conditions at the top and bottom of the plate. The higher-order shear deformation theories (HSDT) (Reddy 1984, 2000, Ren 1986, Touratier 1991, Soldatos 1992, Xiang *et al.* 2009, Akavci 2010, Grover *et al.* 2013, Karama *et al.* 2003, Pradyumna and Bandyopadhyay 2008, Ait Atmane *et al.* 2010, Mantari *et al.* 2012, Kar and Panda 2015a, Karand Panda 2016a, b, Mehar and Panda 2017a, b, Singh and Panda 2017,

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Kolahchi *et al.* 2017, Mehar *et al.* 2017b, Sekkal *et al.* 2017a, Katariya *et al.* 2017b, Hirwani *et al.* 2018) account for shear deformation effects and satisfy the equilibrium conditions at the top and bottom surfaces of the plate without requiring any shear correction factors. Shear deformation models are also employed to study the mechanical response of nanocomposite structures (Kolahchi and Moniri Bidgoli 2016, Madani *et al.* 2016, Kolahchi *et al.* 2016a, b, Arani and Kolahchi 2016, Bilouei *et al.* 2016, Kolahchi *et al.* 2017a, b, c, Kolahchi and Cheraghabak 2017, Kolahchi *et al.* 2017a, b, c, Zamanian *et al.* 2017, Kolahchi 2017, Shokravi 2017a, b, c, d, Besseghier *et al.* 2017, Hajmohammad *et al.* 2017, 2018b, c, d, e, Zarei *et al.* 2017, Amnieh *et al.* 2018, Golabchi *et al.* 2018). In a number of recent articles, a new refined and robust plate theory for bending response and vibration of simply supported FGM plate with only four unknown functions has been developed (Bourada *et al.* 2012, Bachir Bouiadjra *et al.* 2012, Tounsi *et al.* 2013, Kettaf *et al.* 2013, Ait Amar Meziiane *et al.* 2014, Ahmed, 2014, Kolahchi *et al.* 2015, Ait Yahia *et al.* 2015, Kar and Panda, 2015b, Belkorissat *et al.* 2015, Bourada *et al.* 2016, Bouderra *et al.* 2016, Bellifa *et al.* 2016, El-Haina *et al.* 2017, Benadouda *et al.* 2017, Fakhar and Kolahchi 2018, Attia *et al.* 2018, Karami *et al.* 2018f, Fourn *et al.* 2018, Bakhadda *et al.* 2018). Recently, review on refined plate theories can be documented in (Bouderra *et al.* 2013, Tounsi *et al.* 2013, Zidi *et al.* 2014, Larbi Chaht *et al.* 2015, Mahi *et al.* 2015, Karami and Janghorban 2016, Boukhari *et al.* 2016, Bousahla *et al.* 2016, Bounouara *et al.* 2016, Houari *et al.* 2016, Fahsi *et al.* 2017, Khetir *et al.* 2017, Klouche *et al.* 2017, Bellifa *et al.* 2017a, b, Menasria *et al.* 2017, Hachemi *et al.* 2017, Zidi *et al.* 2017, Aldousari 2017, Chikh *et al.* 2017, Shahsavari *et al.* 2018a, Kaci *et al.* 2018, Zine *et al.* 2018, Mokhtar *et al.* 2018, Meksi *et al.* 2019). The thickness stretching effect is neglected in the most of shear deformation theories by considering the transverse displacement as constant. So, this effect should be taken into consideration especially for thick FGM plates (Carrera *et al.* 2011, Sekkal *et al.* 2017b, Younsi *et al.* 2018). To overcome this problem, some quasi-3D theories presented in the literature are developed. Carrera *et al.* (2011) evaluated the effect of thickness stretching in plate/shell structures made by materials which are FGM in the thickness directions. Mantari and Guedes Soares (2013) conducted static analysis of thick FG plates using a novel trigonometric higher-order theory in which stretching effect was considered. Belabed *et al.* (2014) presented an efficient and simple higher order shear and normal deformation theory for FGM plates. Bousahla *et al.* (2014) developed a novel higher order shear and normal deformation theory based on neutral surface position for bending analysis of advanced composite plates. Akavci and Tanrikulu (2015) presented two dimensional (2D) and quasi three-dimensional (quasi-3D) shear deformation theories for static and free vibration analysis of single-layer functionally graded (FG) plates using a new hyperbolic shape function. Draiche *et al.* (2016) presented a refined theory with stretching effect for the flexure analysis of laminated composite plates. Benahmed *et al.* (2017) presented a novel quasi-3D hyperbolic shear deformation theory for functionally graded thick rectangular plates resting on elastic foundation. Shahsavari *et al.* (2018b) proposed a novel quasi-3D hyperbolic theory for free vibration of FG

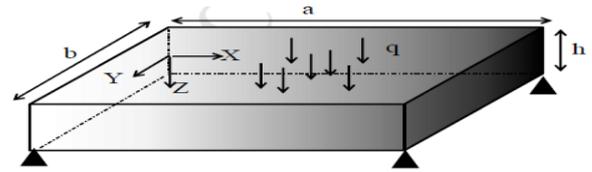


Fig. 1 Schematic representation of FGM plate with distribution of volume fraction along  $x$

plates with porosities resting on Winkler/Pasternak/Kerr foundation. Benchohra *et al.* (2018) employed a new quasi-3D sinusoidal shear deformation theory for FG plates. Abualnour *et al.* (2018) proposed a novel quasi-3D trigonometric plate theory for free vibration analysis of advanced composite plates. Hirwani and Panda (2019) presented nonlinear finite element solutions of thermoelastic deflection and stress responses of internally damaged curved panel structure and two different polynomial types of kinematic theories including the through-thickness stretching effect are employed. Katariya *et al.* (2018) used higher-order kinematic theory including the stretching term effect to study the nonlinear deflection and stress of skew sandwich shell panel. Shahsavari *et al.* (2018c) studied the shear buckling of porous nanoplates using a new size-dependent quasi-3D shear deformation theory. Galerkin method is employed to find the shear buckling forces. Bouhadra *et al.* (2018) proposed an improved HSDT accounting for effect of thickness stretching in advanced composite plates. Zaoui *et al.* (2019) developed a new 2D and quasi-3D shear deformation theories for free vibration of FG plates on elastic foundations.

It is seen from the above literature analysis that most of the previous studies are related to FG structures whose material properties vary through the thickness only. Thus, there have been a very little number of studies related to FG structures with property variation throughout the length so far.

A novel method has been developed by Amirpour *et al.* (2016) to analyze the elastic deformation of functionally graded thick plates with in-plane stiffness variation using higher order shear deformation theory. For design and analysis of thick FG plates with in-plane stiffness variation, the development of simple theory for bending analysis of thick FG plates with in-plane property variation, including stretching effects, is very relevant and represents the main objective of the present article.

The aim of this work is to analyze bending behavior of thick FG plates with in-plane variation of stiffness (variation of stiffness through the length of the plate) using the higher shear deformation theory including stretching of the thickness. The main challenge in developing accurate models for in-plane property variation compared to the through-the-thickness property variation is that the variation of Young's modulus (material stiffness) through the length ( $x$ ) leads to mathematically complex five simultaneous governing equations. The solutions of which become relatively difficult as several parameters vary with the length ( $x$ ). The transverse displacement is apportioned into three components: bending, shear and thickness stretching. Equations of motion are derived from the principle of

virtual displacements. The accuracy of obtained solutions is verified by comparing the present results with those predicted by solutions available in the literature.

## 2. Theoretical developments

Consider a simply supported rectangular FG plate with the length  $a$  width  $b$ , and thickness  $h$ . The  $x$ -,  $y$ -, and  $z$ -coordinates are taken along the length, width, and height of the plate, respectively, as shown in Fig. 1. The formulation is limited to linear elastic material behavior. The FG plate is isotropic with its material properties vary smoothly through the length of the plate.

### 2.1 Displacement field and strains

The formulation is limited to linear elastic material behavior. The displacement fields of various shear deformation theories are chosen based on following assumptions: (1) The transverse displacement is partitioned into bending and shear and stretching components, (2) the in-plane displacements are partitioned into extension, bending and shear components, (3) the bending parts of the in-plane displacements are similar to those given by the classical plate theory (CPT) and the shear component of axial displacement gives rise to the higher-order variation of shear strain and hence to shear stress through the thickness of the plate in such a way that shear stress vanishes on the top and bottom surfaces. Based on these assumptions, the displacement fields of various higher-orders shear deformation theories are given in a general form as (Hebali *et al.* 2014, Hamidi *et al.* 2015, Bennoun *et al.* 2016, Bouafia *et al.* 2017, Shahsavari *et al.* 2018b)

$$\begin{aligned} u(x, y, z) &= u_0(x, y) - z \frac{\partial w_b}{\partial x} - f(z) \frac{\partial w_s}{\partial x} \\ v(x, y, z) &= v_0(x, y) - z \frac{\partial w_b}{\partial y} - f(z) \frac{\partial w_s}{\partial y} \\ w(x, y, z) &= w_b(x, y) + w_s(x, y) + g(z)\varphi(x, y) \end{aligned} \tag{1a}$$

The shape function is given as (Zenkour 2013, Bourada *et al.* 2015)

$$f(z) = z - \left[ h \sinh\left(\frac{z}{h}\right) - \frac{4}{3} \left(\frac{z^3}{h^2}\right) \cosh\frac{1}{2} \right] \tag{1b}$$

where  $u_0$  and  $v_0$  denote the displacements along the  $x$  and  $y$  coordinate directions of a point on the mid-plane of the plate,  $w_b$  and  $w_s$  are the bending and shear components of the transverse displacement, respectively, and the additional displacement  $\varphi$  accounts for the stretching effect.  $f(z)$  is a shape function determining the distribution of the transverse shear strain and shear stress through the thickness of the plate. The shape functions  $f(z)$  are chosen to satisfy the stress-free boundary conditions on the top and bottom surfaces of the plate, thus a shear correction factor is not required. Note that  $g(z)=0$  is required for 2D analysis.

The nonzero linear strains associated with the

displacement field in Eq. (1) are

$$\begin{aligned} \begin{Bmatrix} \varepsilon_x \\ \varepsilon_y \\ \gamma_{xy} \end{Bmatrix} &= \begin{Bmatrix} \varepsilon_x^0 \\ \varepsilon_y^0 \\ \gamma_{xy}^0 \end{Bmatrix} + z \begin{Bmatrix} k_x^b \\ k_y^b \\ k_{xy}^b \end{Bmatrix} + f(z) \begin{Bmatrix} k_x^s \\ k_y^s \\ k_{xy}^s \end{Bmatrix} \\ \begin{Bmatrix} \gamma_{yz} \\ \gamma_{xz} \end{Bmatrix} &= g(z) \begin{Bmatrix} \gamma_{yz}^0 \\ \gamma_{xz}^0 \end{Bmatrix}, \\ \varepsilon_z &= g'(z)\varepsilon_z^0, \end{aligned} \tag{2}$$

Where

$$\begin{aligned} \begin{Bmatrix} \varepsilon_x^0 \\ \varepsilon_y^0 \\ \gamma_{xy}^0 \end{Bmatrix} &= \begin{Bmatrix} \frac{\partial u_0}{\partial x} \\ \frac{\partial v_0}{\partial y} \\ \frac{\partial u_0}{\partial y} + \frac{\partial v_0}{\partial x} \end{Bmatrix}, \quad \begin{Bmatrix} k_x^b \\ k_y^b \\ k_{xy}^b \end{Bmatrix} = \begin{Bmatrix} -\frac{\partial^2 w_b}{\partial x^2} \\ -\frac{\partial^2 w_b}{\partial y^2} \\ -2\frac{\partial^2 w_b}{\partial x \partial y} \end{Bmatrix}, \\ \begin{Bmatrix} k_x^s \\ k_y^s \\ k_{xy}^s \end{Bmatrix} &= \begin{Bmatrix} -\frac{\partial^2 w_s}{\partial x^2} \\ -\frac{\partial^2 w_s}{\partial y^2} \\ -2\frac{\partial^2 w_s}{\partial x \partial y} \end{Bmatrix}, \\ \begin{Bmatrix} \gamma_{yz}^0 \\ \gamma_{xz}^0 \end{Bmatrix} &= \begin{Bmatrix} \frac{\partial w_s}{\partial y} + \frac{\partial \varphi}{\partial y} \\ \frac{\partial w_s}{\partial x} + \frac{\partial \varphi}{\partial x} \end{Bmatrix}, \quad \varepsilon_z^0 = \varphi(x, y) \end{aligned} \tag{3}$$

and

$$g(z) = 1 - f'(z) \tag{4}$$

### 2.2 Constitutive relations

The material is assumed to be elastic and inhomogeneous, and the material stiffness vary continuously through the length of the plate and obey a simple power-law distribution of volume fraction of constituents as given by (Amirpour *et al.* 2016)

$$V_c(x) = \left(\frac{x}{a}\right)^p \tag{5}$$

Where  $p$  is a parameter that governs the material variation profile through the length of the plate. Since the effect of variation of Poisson's ratio on the response of FG plates is very small (Yang 2005, Kitipornchai 2006), this material parameter is assumed to be constant, and the Young's modulus is considered to be variable, and can be

determined by the rule of mixture as (Tornabene 2009)

$$E(x) = \left(\frac{x}{a}\right)^p (E_c - E_m) + E_m \tag{6}$$

The linear constitutive relations of a FG plate can be written as

$$\begin{Bmatrix} \sigma_x \\ \sigma_y \\ \sigma_z \\ \tau_{yz} \\ \tau_{xz} \\ \tau_{xy} \end{Bmatrix} = \begin{bmatrix} C_{11} & C_{12} & C_{13} & 0 & 0 & 0 \\ C_{12} & C_{22} & C_{23} & 0 & 0 & 0 \\ C_{13} & C_{23} & C_{33} & 0 & 0 & 0 \\ 0 & 0 & 0 & C_{44} & 0 & 0 \\ 0 & 0 & 0 & 0 & C_{55} & 0 \\ 0 & 0 & 0 & 0 & 0 & C_{66} \end{bmatrix} \begin{Bmatrix} \epsilon_x \\ \epsilon_y \\ \epsilon_z \\ \gamma_{yz} \\ \gamma_{xz} \\ \gamma_{xy} \end{Bmatrix} \tag{7}$$

where  $(\sigma_x, \sigma_y, \sigma_z, \tau_{yz}, \tau_{xz}, \tau_{xy})$  and  $(\epsilon_x, \epsilon_y, \epsilon_z, \gamma_{yz}, \gamma_{xz}, \gamma_{xy})$  are the stress and strain components, respectively. Using the material properties defined in Eq. (6), stiffness coefficients,  $C_{ij}$ , can be expressed as

$$C_{11} = C_{22} = C_{33} = \frac{1-\nu}{(1+\nu)(1-2\nu)} E(x), \tag{8a}$$

$$C_{12} = C_{13} = C_{23} = \frac{\nu}{(1+\nu)(1-2\nu)} E(x) \tag{8b}$$

$$C_{44} = C_{55} = C_{66} = \frac{E(x)}{2(1+\nu)}, \tag{8c}$$

If the  $\epsilon_z=0$ , then stiffness coefficients,  $C_{ij}$ , are,

$$C_{11} = C_{22} = \frac{E(x)}{1-\nu^2(x)}, \tag{9a}$$

$$C_{12} = \nu C_{11}, \tag{9b}$$

$$C_{44} = C_{55} = C_{66} = \frac{E(x)}{2(1+\nu)}, \tag{9c}$$

### 2.3 Equations of motion

The governing equations of the present theory are derived from the Principle of Virtual Displacements. The internal virtual work is initially formulated as follows

$$\delta U + \delta V = 0 \tag{10}$$

Where  $\delta U, \delta V$  are the variation of the strain energy, the external work by applied forces, respectively, and are described in the following equations

$$\begin{aligned} \delta U &= \int_V \left[ \sigma_x \delta \epsilon_x + \sigma_y \delta \epsilon_y + \sigma_z \delta \epsilon_z + \tau_{xy} \delta \gamma_{xy} + \tau_{yz} \delta \gamma_{yz} + \tau_{xz} \delta \gamma_{xz} \right] dV \\ &= \int_A \left[ N_x \delta \epsilon_x^0 + N_y \delta \epsilon_y^0 + N_z \delta \epsilon_z^0 + N_{xy} \delta \gamma_{xy}^0 + M_x^b \delta \kappa_x^b + M_y^b \delta \kappa_y^b + M_{xy}^b \delta \kappa_{xy}^b \right. \\ &\quad \left. + M_x^s \delta \kappa_x^s + M_y^s \delta \kappa_y^s + M_{xy}^s \delta \kappa_{xy}^s + S_{xz}^s \delta \gamma_{xz}^s + S_{yz}^s \delta \gamma_{yz}^s \right] dA = 0 \end{aligned} \tag{11}$$

Where  $A$  is the top surface and the stress resultants  $N, M$ , and  $S$  are defined by

$$\begin{Bmatrix} N_x & N_y & N_{xy} \\ M_x^b & M_y^b & M_{xy}^b \\ M_x^s & M_y^s & M_{xy}^s \end{Bmatrix} = \int_{-h/2}^{h/2} (\sigma_x, \sigma_y, \tau_{xy}) \begin{Bmatrix} 1 \\ z \\ f(z) \end{Bmatrix} dz \tag{12a}$$

$$(S_{xz}^s, S_{yz}^s) = \int_{-h/2}^{h/2} (\tau_{xz}, \tau_{yz}) g(z) dz \quad N_z = \int_{-h/2}^{h/2} \sigma_z g'(z) dz \tag{12b}$$

Substituting Eq. (7) into Eq. (12) and integrating through the thickness of the plate, the stress resultants are given as

$$M_x^b = B_{12} d_1 u_0 + B_{12} d_2 v_0 - D_{11} d_{11} w_b - D_{12} d_{22} w_b - D_{11}^s d_{11} w_s - D_{12}^s d_{22} w_s + L^2 \phi \tag{13b}$$

$$M_y^b = B_{12} d_1 u_0 + B_{22} d_2 v_0 - D_{12} d_{11} w_b - D_{22} d_{22} w_b - D_{12}^s d_{11} w_s - D_{22}^s d_{22} w_s + L^2 \phi \tag{13b}$$

$$M_{xy}^b = B_{66} d_2 u_0 + B_{66} d_1 v_0 - 2D_{66} d_{12} w_b - 2d_1 D_{66}^s d_{12} w_s \tag{13c}$$

$$M_x^s = B_{11}^s d_1 u_0 + B_{12}^s d_2 v_0 - D_{11}^s d_{11} w_b - D_{12}^s d_{22} w_b - H_{11}^s d_{11} w_s - H_{12}^s d_{22} w_s + L^3 \phi \tag{13c}$$

$$M_y^s = B_{12}^s d_1 u_0 + B_{22}^s d_2 v_0 - D_{12}^s d_{11} w_b - D_{22}^s d_{22} w_b - H_{12}^s d_{11} w_s - H_{22}^s d_{22} w_s + L^3 \phi \tag{13c}$$

$$M_{xy}^s = B_{66}^s d_2 u_0 + B_{66}^s d_1 v_0 - 2D_{66}^s d_{12} w_b - 2H_{66}^s d_{12} w_s \tag{13c}$$

$$S_{xz} = A_{55}^s d_1 w_s + A_{55}^s d_1 \phi \tag{13d}$$

$$S_{yz} = A_{44}^s d_2 w_s + A_{55}^s d_2 \phi \tag{13d}$$

The variation of work done by the applied loads can be expressed as

$$\delta V = - \int_A q (\delta w_b + \delta w_s + \delta w_{sy}) dA \tag{14}$$

Substituting the expressions for  $\delta U$  and  $\delta V$  from Eqs. (11) and (14) into Eq. (10) and integrating by parts, and then collecting the coefficients of  $\delta u_0, \delta v_0, \delta w_b, \delta w_s$  and  $\delta \phi$ , the following governing equations of the FG plate with the variation of material property through the length are obtained as follows

$$\begin{aligned} \delta u_0 : \frac{\partial N_x}{\partial x} + \frac{\partial N_{xy}}{\partial y} &= 0 \\ \delta v_0 : \frac{\partial N_y}{\partial y} + \frac{\partial N_{xy}}{\partial x} &= 0 \\ \delta w_b : \frac{\partial^2 M_x^b}{\partial x^2} + \frac{\partial^2 M_y^b}{\partial y^2} + 2 \frac{\partial^2 M_{xy}^b}{\partial x \partial y} + q &= 0 \\ \delta w_s : \frac{\partial^2 M_x^s}{\partial x^2} + \frac{\partial^2 M_y^s}{\partial y^2} + 2 \frac{\partial^2 M_{xy}^s}{\partial x \partial y} + \frac{\partial S_{xz}^s}{\partial x} + \frac{\partial S_{yz}^s}{\partial y} + q &= 0 \\ \delta \phi : \frac{\partial S_{xz}^s}{\partial x} + \frac{\partial S_{yz}^s}{\partial y} + qg(z) - N_z &= 0 \end{aligned} \tag{15}$$

Introducing Eq. (13) into Eq. (15), equations of motion for FG plate can be obtained as follows

$$A_1 d_1 u_0 + d_1 A_1 d_1 u_0 + A_{66} d_{22} u_0 + (A_2 + A_{66}) d_{12} v_0 + d_1 A_2 d_2 v_0 - d_{11} w_b B_{11} - d_{11} w_s B_{11} - d_{11} w_b B_{11}^s - d_{11} w_s B_{11}^s - d_{12} w_b B_{12} - d_{12} w_s B_{12} - d_{12} w_b B_{12}^s - d_{12} w_s B_{12}^s - 2d_{22} w_b B_{22} - 2d_{22} w_s B_{22} - 2d_{22} w_b B_{22}^s - 2d_{22} w_s B_{22}^s + L d_1 \phi + d_1 L \phi = 0 \tag{16a}$$

$$A_2 d_2 u_0 + A_{22} d_{22} v_0 + d_1 A_{66} d_{12} v_0 + A_{66} d_{11} v_0 + d_1 A_{66} d_2 u_0 + A_{66} d_{12} u_0 - 2d_{11} w_b B_{66} - 2d_{12} w_b B_{66} - 2d_{12} w_s B_{66}^s - 2d_{12} w_b B_{66} - d_{11} w_b B_{66} - d_{11} w_s B_{66}^s - d_{12} w_b B_{66} - d_{12} w_s B_{66}^s - 2d_{22} w_b B_{22} - d_{22} w_s B_{22} - 2d_{22} w_b B_{22}^s - 2d_{22} w_s B_{22}^s + L d_2 \phi = 0 \tag{16b}$$

$$\begin{aligned} -d_{11} D_{11} d_{11} w_b - 2d_{11} D_{11} d_{11} w_b - D_{11} d_{11} w_b - d_{11} D_{12} d_{22} w_b - 2d_{12} D_{12} d_{12} w_b - D_{12} d_{12} w_b \\ - d_{11} D_{12} d_{11} w_s - 2d_{12} D_{12} d_{11} w_s - D_{12} d_{11} w_s - d_{11} D_{12} d_{22} w_s - 2d_{12} D_{12} d_{22} w_s - D_{12} d_{22} w_s \\ + d_{11} L \phi + 2d_{12} L \phi + L d_1 \phi - D_{12} d_{12} w_b - D_{22} d_{22} w_b - D_{12}^s d_{12} w_s - D_{22}^s d_{22} w_s \\ + L^2 d_{22} \phi - 4d_1 D_{66} d_{12} w_b - 4D_{66} d_{12} w_b - 4d_1 D_{66}^s d_{12} w_s - 4D_{66}^s d_{12} w_s + B_{12} d_{12} v_0 \\ + d_{11} B_{12} d_{22} v_0 + 2d_1 B_{12} d_{22} v_0 + B_{11} d_{11} u_0 + d_{11} B_{11} d_1 u_0 + 2B_{66} d_{12} v_0 + 2d_1 B_{66} d_{12} v_0 \\ + 2B_{66} d_{12} u_0 + 2d_1 B_{66} d_{22} u_0 + 2d_1 B_{12} d_{12} v_0 + B_{12} d_{12} u_0 + B_{22} d_{22} v_0 + q = 0 \end{aligned} \tag{16c}$$

$$\begin{aligned}
 & -d_1 D_1^x d_{11} w_b - 2d_1 D_1^x d_{11} w_b - D_1^x d_{111} w_b - d_1 D_1^x d_{22} w_b - 2d_1 D_1^x d_{22} w_b - D_1^x d_{222} w_b \\
 & + d_1 L^2 \phi + 2d_1 L^2 d_1 \phi + L^2 d_{22} \phi - d_1 H_1^x d_{11} w_s - 2d_1 H_1^x d_{11} w_s - H_1^x d_{111} w_s \\
 & - d_1 H_1^x d_{22} w_s - 2d_1 H_1^x d_{22} w_s - H_1^x d_{112} w_s - H_1^x d_{112} w_s - D_1^x d_{112} w_b - D_1^x d_{222} w_b \\
 & - H_1^x d_{122} w_s - H_1^x d_{222} w_s - 4d_1 D_{66}^x d_{122} w_b - 4D_{66}^x d_{122} w_b - 4d_1 H_{66}^x d_{122} w_s - 4H_{66}^x d_{122} w_s \\
 & + d_1 A_{33}^x d_{11} w_s + A_{33}^x d_{11} w_s + d_1 A_{33}^x d_1 \phi + A_{33}^x d_1 \phi + A_{33}^x d_{22} w_s + A_{33}^x d_{22} \phi + 2B_{66}^x d_{12} v_0 \\
 & + 2d_1 B_{66}^x d_{22} v_0 + B_{66}^x d_{222} v_0 + 2d_1 B_1^x d_{11} u_0 + B_1^x d_{11} u_0 + 2d_1 B_1^x d_{11} u_0 + d_1 B_1^x d_{11} u_0 \\
 & + 2d_1 B_{66}^x d_{12} v_0 + 2B_{66}^x d_{122} v_0 + B_1^x d_{122} v_0 + B_1^x d_{112} v_0 + 2d_1 B_1^x d_{11} v_0 + q = 0
 \end{aligned} \tag{16d}$$

$$\begin{aligned}
 & d_1 A_{55}^x d_1 w_s + A_{55}^x d_{11} w_s + d_1 A_{55}^x d_1 \phi + A_{55}^x d_{11} \phi + A_{55}^x d_{22} w_s + A_{55}^x d_{22} \phi + L^2 (d_{11} w_b + d_{22} w_b) \\
 & + L^2 (d_{11} w_s + d_{22} w_s) - L^2 \phi = 0
 \end{aligned} \tag{16e}$$

Where  $d_{ij}$ ,  $d_{ijl}$  and  $d_{ijlm}$  are the following differential operators

$$\begin{aligned}
 d_{ij} &= \frac{\partial^2}{\partial x_i \partial x_j}, \quad d_{ijl} = \frac{\partial^3}{\partial x_i \partial x_j \partial x_l}, \quad d_{ijlm} = \frac{\partial^4}{\partial x_i \partial x_j \partial x_l \partial x_m}, \\
 d_i &= \frac{\partial}{\partial x_i}, \quad (i, j, l, m = 1, 2).
 \end{aligned} \tag{17}$$

And stiffness components are given as

$$(A_{ij}, B_{ij}, D_{ij}, B_{ij}^x, D_{ij}^x, H_{ij}^x) = \int_{-h/2}^{h/2} C_{ij} (1, z, z^2, f(z), z f(z), f^2(z)) dz \tag{18a}$$

$$A_{44}^s = A_{55}^s = \int_{-h/2}^{h/2} C_{44} [g'(z)]^2 dz, \tag{18b}$$

$$L^1 = \int_{-h/2}^{h/2} C_{13} g'(z) dz; \quad L^2 = \int_{-h/2}^{h/2} C_{13} z g'(z) dz; \quad L^3 = \int_{-h/2}^{h/2} C_{13} g'(z) f(z) dz; \quad L^4 = \int_{-h/2}^{h/2} C_{33} [g'(z)]^2 dz \tag{18c}$$

### 3. Analytical solutions

The exact solution of Eq. (16) for the FGM plate under various boundary conditions can be constructed. The boundary conditions for an arbitrary edge with simply supported conditions are:

• Simply supported (S)

$$\begin{aligned}
 v_0 = w_b = \partial w_b / \partial y = w_s = \partial w_s / \partial y = 0 & \quad \text{at } x = 0, a \\
 u_0 = w_b = \partial w_b / \partial x = w_s = \partial w_s / \partial x = 0 & \quad \text{at } y = 0, b
 \end{aligned} \tag{19}$$

The following representation for the displacement quantities, that satisfy the above boundary conditions, is appropriate in the case of our problem

$$\begin{Bmatrix} u_0 \\ v_0 \\ w_b \\ w_s \\ \phi \end{Bmatrix} = \sum_{m=1}^{\infty} \sum_{n=1}^{\infty} \begin{Bmatrix} U_{mn} \frac{\partial X_m(x)}{\partial x} Y_n(y) \\ V_{mn} X_m(x) \frac{\partial Y_n(y)}{\partial y} \\ W_{bmn} X_m(x) Y_n(y) \\ W_{smn} X_m(x) Y_n(y) \\ \varphi_{mn} X_m(x) Y_n(y) \end{Bmatrix} \tag{20}$$

For simply supported boundary conditions

$$X_m(x) = \sin(\lambda x) \quad ; \quad Y_n(y) = \sin(\mu y) \tag{21}$$

The transverse load  $q$  is also expanded in the double-Fourier sine series as follows

$$q(x, y) = \sum_{m=1}^{\infty} \sum_{n=1}^{\infty} Q_{mn} \sin(\lambda x) \sin(\mu y) \tag{22}$$

The coefficients  $Q_{mn}$  are given below for some typical loads (Zenkour 2006)

$$Q_{mn} = \frac{4}{ab} \int_0^a \int_0^b q(x, y) \sin(\lambda x) \sin(\mu y) dx dy = \begin{cases} q_0 & \text{for sinusoidal load} \\ \frac{16q_0}{mn\pi^2} & \text{for uniformly distribute d load} \end{cases} \tag{23}$$

where  $U_{mn}$ ,  $V_{mn}$ ,  $W_{bmn}$ ,  $W_{smn}$  and  $\varphi_{mn}$  are arbitrary parameters to be determined. The functions  $X_m(x)$  and  $Y_n(y)$  are suggested here to satisfy at least the geometric boundary conditions given in Eqs. (21) and represent approximate shapes of the deflected surface of the plate. Noting that  $\lambda = m\pi/a$  and  $\mu = n\pi/b$ .

By substituting Eqs. (20) and (23) into Eq. (16), a set of algebraic equations can be obtained as

$$\begin{Bmatrix} a_{11} & a_{12} & a_{13} & a_{14} & a_{15} \\ a_{21} & a_{22} & a_{23} & a_{24} & a_{25} \\ a_{31} & a_{32} & a_{33} & a_{34} & a_{35} \\ a_{41} & a_{42} & a_{43} & a_{44} & a_{45} \\ a_{51} & a_{52} & a_{53} & a_{54} & a_{55} \end{Bmatrix} \begin{Bmatrix} U_{mn} \\ V_{mn} \\ W_{bmn} \\ W_{smn} \\ \varphi_{mn} \end{Bmatrix} = \begin{Bmatrix} 0 \\ 0 \\ q_0 \\ q_0 \\ 0 \end{Bmatrix} \tag{24}$$

Where the coefficients of the above matrix are given as follows

$$\begin{aligned}
 a_{11} &= \int_0^a \int_0^b (A_1 \alpha_2 + \alpha_3 d_1 A_1 + \alpha_4 A_3) \alpha_5 dx dy \\
 a_{12} &= \int_0^a \int_0^b (A_2 \alpha_5 + \alpha_3 d_1 A_2 + \alpha_4 A_3) \alpha_6 dx dy \\
 a_{21} &= \int_0^a \int_0^b (A_3 \alpha_{10} + \alpha_2 d_1 A_3 + \alpha_{10} A_2) \alpha_2 dx dy \\
 a_{23} &= \int_0^a \int_0^b (-B_1 \alpha_{12} - \alpha_5 d_1 B_1 - \alpha_5 B_{12} - \alpha_5 d_1 B_{12} - 2\alpha_5 B_{66}) \alpha_5 dx dy \\
 a_{31} &= \int_0^a \int_0^b (B_1 \alpha_{11} + 2\alpha_4 d_1 B_{66} + 2\alpha_1 d_1 B_{66} + \alpha_4 d_1 B_{11} + 2\alpha_{12} d_1 B_{11} + \alpha_{11} B_{11}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-B_1^x \alpha_{12} - d_1 B_1^x \alpha_5 - B_1^x \alpha_4 - d_1 B_1^x \alpha_5 - 2B_{66}^x \alpha_5) \alpha_5 dx dy \\
 a_{41} &= \int_0^a \int_0^b (2d_1 B_{66}^x \alpha_5 + B_{66}^x \alpha_{11} + d_{11} B_1^x \alpha_5 + 2B_{66}^x \alpha_{11} + 2d_1 B_1^x \alpha_{12} + B_1^x \alpha_{12}) \alpha_1 dx dy \\
 a_{45} &= \int_0^a \int_0^b (L^2 \alpha_5 + d_1 L^2 \alpha_1) \alpha_5 dx dy \\
 a_{51} &= \int_0^a \int_0^b (-L^2 \alpha_5) \alpha_5 dx dy \\
 a_{52} &= \int_0^a \int_0^b (A_{66} \alpha_{10} + \alpha_7 d_1 A_{66} + \alpha_4 A_{22}) \alpha_2 dx dy \\
 a_{53} &= \int_0^a \int_0^b (-\alpha_4 B_{22} - \alpha_{10} B_{12} - 2\alpha_4 d_1 B_{66} - 2\alpha_{10} B_{66}) \alpha_2 dx dy \\
 a_{54} &= \int_0^a \int_0^b (2\alpha_5 d_1 B_{12} + \alpha_2 B_{22} + 2\alpha_5 d_1 B_{66} + 2\alpha_1 d_1 B_{66} + \alpha_5 d_1 B_{12} + \alpha_{11} B_{12}) \alpha_1 dx dy \\
 a_{54} &= \int_0^a \int_0^b (-B_{22}^x \alpha_4 - B_{12}^x \alpha_{10} - 2d_1 B_{66}^x \alpha_5 - 2B_{66}^x \alpha_{10}) \alpha_2 dx dy \\
 a_{54} &= \int_0^a \int_0^b (2d_1 B_{12}^x \alpha_5 + d_{11} B_1^x \alpha_5 + B_{12}^x \alpha_{11} + 2B_{66}^x \alpha_{11} + 2d_1 B_{66}^x \alpha_5 + B_{22}^x \alpha_1) \alpha_1 dx dy \\
 a_{55} &= \int_0^a \int_0^b (L^2 \alpha_5) \alpha_5 dx dy \\
 a_{52} &= \int_0^a \int_0^b (-L^2 \alpha_5) \alpha_5 dx dy \\
 a_{33} &= \int_0^a \int_0^b (-\alpha_{13} D_{11} - 2d_1 D_{11} \alpha_{12} - 2d_1 D_{12} \alpha_8 - 2D_{12} \alpha_{11} - d_{11} D_{12} \alpha_5 - D_{22} \alpha_5 - d_{11} D_{11} \alpha_5 \\
 & - 4d_1 D_{66} \alpha_8 - 4D_{66} \alpha_{11}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1 dx dy \\
 a_{34} &= \int_0^a \int_0^b (-\alpha_5 D_{22} - 2d_1 D_{11} \alpha_{12} - 2D_{12}^x \alpha_{11} - d_{11} D_{12}^x \alpha_5 - 2d_1 D_{12}^x \alpha_8 - 4d_1 D_{66}^x \alpha_8 - 4D_{66}^x \alpha_{11} \\
 & - d_{11} D_{11}^x \alpha_5 - D_{11}^x \alpha_{13}) \alpha_1$$

Table 1 Material properties used in the functionally graded plates

Material	Properties	
	$E$	$\nu$
Aluminium (Al)	70	0.3
Alumina (Al <sub>2</sub> O <sub>3</sub> )	380	0.3

Table 2 Comparison of the deflection and stress components of a simply supported homogeneous square plate ( $a/h = 10$ )

Loading	Theory	$\epsilon_z$	$\bar{w}(0)$	$\bar{\sigma}_x\left(\frac{h}{2}\right)$	$\bar{\sigma}_y\left(\frac{h}{3}\right)$	$\bar{\tau}_{xz}(0)$	$\bar{\tau}_{yz}\left(\frac{h}{6}\right)$	$\bar{\tau}_{xy}\left(-\frac{h}{3}\right)$
Zenkour (2006)	Present	=0	0.4665	2.8932	1.9103	0.5114	0.4429	1.2850
	Present	=0	0.4665	2.9809	1.9103	0.4988	0.4363	1.2857
UL	Present	≠0	0.4635	2.9981	1.8925	0.4782	0.4315	1.2578
	Present	≠0	0.4639	2.9064	1.9153	0.4924	0.4376	1.2557
SL	Present	=0	0.4665	2.8921	1.9106	0.4958	0.4406	1.2857
	Present	=0	0.2960	1.9955	1.3121	0.2462	0.2132	0.7065
SL	Present	=0	0.2960	1.9943	1.3123	0.2387	0.2121	0.7066
	Present	≠0	0.2942	2.0075	1.3173	0.2386	0.2120	0.7010

Table 3 The non-dimensional displacement and stress components of an Al/Al<sub>2</sub>O<sub>3</sub> FG square plate subjected to sinusoidal load ( $a/h = 10$ )

$p$	$\epsilon_z$	$\bar{w}(0)$	$\bar{\sigma}_x\left(\frac{h}{2}\right)$	$\bar{\sigma}_y\left(\frac{h}{3}\right)$	$\bar{\tau}_{xz}(0)$	$\bar{\tau}_{yz}\left(\frac{h}{6}\right)$	$\bar{\tau}_{xy}\left(-\frac{h}{3}\right)$
1	≠0	0.4965	2.0076	1.3172	0.0742	0.2120	0.2180
	=0	0.5000	1.9943	1.3123	0.0742	0.2121	0.2198
2	≠0	0.6621	1.7570	1.1535	0.0964	0.1805	0.2910
	=0	0.6662	1.7441	1.1481	0.0964	0.1805	0.2934
4	≠0	0.8987	1.4490	0.9518	0.1266	0.1436	0.3958
	=0	0.9043	1.4365	0.9460	0.1266	0.1436	0.3989
10	≠0	1.2173	1.5449	1.0149	0.1701	0.1518	0.5363
	=0	1.2248	1.5311	1.0084	0.1702	0.1519	0.5406

With

$$\begin{aligned}
 (\alpha_1, \alpha_2, \alpha_3, \alpha_4, \alpha_5) &= X_m(Y_n, d_2Y_n, d_{22}Y_n, d_{222}Y_n, d_{2222}Y_n) \\
 (\alpha_6, \alpha_7, \alpha_8) &= d_1X_m(Y_n, d_2Y_n, d_{22}Y_n) \\
 (\alpha_9, \alpha_{10}, \alpha_{11}) &= d_{11}X_m(Y_n, d_2Y_n, d_{22}Y_n) \\
 (\alpha_{12}, \alpha_{13}) &= d_1Y_n(d_{111}X_m, d_{1111}X_m)
 \end{aligned}
 \tag{26}$$

### 4. Numerical results and discussions

In order to verify the accuracy of the present theory in predicting the static response of simply supported FG plates, different examples are solved and compared with the results of various quasi-3D and 2D shear deformation theories. The material properties of the plate are listed in Table 1. The length of the plate is assumed to be 1 m.

The non-dimensional displacement and stress components are

Table 4 The non-dimensional displacement and stress components of an Al/Al<sub>2</sub>O<sub>3</sub> FG square plate subjected to sinusoidal load

$p$	theory	$\epsilon_z$	$\bar{\sigma}_x\left(\frac{h}{2}\right)$			$\bar{w}(0)$		
			$a/h=4$	$a/h=10$	$a/h=100$	$a/h=4$	$a/h=10$	$a/h=100$
1	Present	≠0	0.8686	2.0076	19.7752	0.6189	0.4965	0.4736
	present	=0	0.8365	1.9943	19.7594	0.6394	0.5000	0.4736
4	Present	≠0	0.6318	1.4490	14.2524	1.1044	0.8987	0.8592
	present	=0	0.6009	1.4365	14.2408	1.1421	0.9043	0.8592
10	Present	≠0	0.6745	1.5449	15.1914	1.4934	1.2173	1.1642
	present	=0	0.6402	1.5311	15.1790	1.5445	1.2248	1.1643

Table 6 Non-dimensional stress  $\bar{\sigma}_x\left(\frac{h}{2}\right)$  of an Al/Al<sub>2</sub>O<sub>3</sub> FG plate subjected to sinusoidal distributed load ( $a/h=10$ )

$b/a$	theory	$\epsilon_z$	$p$			
			0	1	4	10
1	Present	≠0	2.0075	2.0076	1.4490	1.5449
	Present	=0	1.9943	1.9944	1.4365	1.5311
2	Present	≠0	4.2148	4.2149	3.2089	3.4458
	Present	=0	4.2071	4.2071	3.2017	3.4379
3	Present	≠0	5.1208	5.1209	4.0779	4.4071
	Present	=0	5.1148	5.1149	4.0732	4.4019

$$\begin{aligned}
 \bar{w} &= \frac{10h^3 E_c}{q_0 a^4} w\left(\frac{a}{2}, \frac{b}{2}, z\right) \\
 \bar{\sigma}_x(z) &= \frac{h}{q_0 a} \sigma_x\left(\frac{a}{2}, \frac{b}{2}, z\right) \\
 \bar{\sigma}_y(z) &= \frac{h}{q_0 a} \sigma_y\left(\frac{a}{2}, \frac{b}{2}, z\right) \\
 \bar{\tau}_{xy}(z) &= \frac{h}{q_0 a} \tau_{xy}(0, 0, z) \quad \bar{\tau}_{xz}(z) = \frac{h}{q_0 a} \tau_{xz}\left(0, \frac{b}{2}, z\right) \\
 \bar{\tau}_{yz}(z) &= \frac{h}{q_0 a} \tau_{yz}\left(\frac{a}{2}, 0, z\right)
 \end{aligned}
 \tag{27}$$

The non-dimensional displacement and stress components for Fig. 6

$$\begin{aligned}
 \bar{u}(z) &= \frac{E_c}{hq_0} u\left(0, \frac{b}{2}, z\right) \quad \bar{w} = \frac{E_c}{hq_0} w\left(\frac{a}{2}, \frac{b}{2}, z\right) \quad \bar{\sigma}_x(z) = \frac{\sigma_x\left(\frac{a}{2}, \frac{b}{2}, z\right)}{q_0} \\
 \bar{\sigma}_z(z) &= \frac{\sigma_z\left(\frac{a}{2}, \frac{b}{2}, z\right)}{q_0} \quad \bar{\tau}_{xy}(z) = \frac{\tau_{xy}(0, 0, z)}{q_0} \\
 \bar{\tau}_{xz}(z) &= \frac{\tau_{xz}\left(0, \frac{b}{2}, z\right)}{q_0}
 \end{aligned}
 \tag{28}$$

In Table 2, the non-dimensional displacement and stresses of a simply supported homogeneous square plate

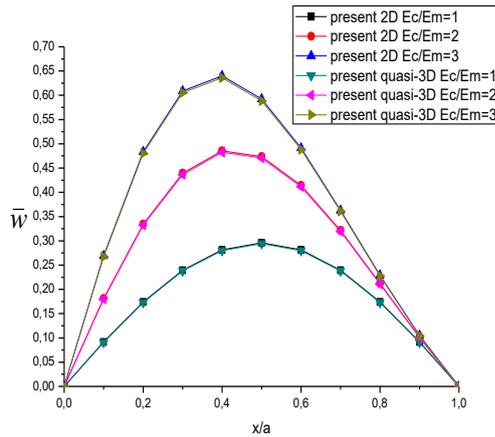


Fig. 2 Deflection of the FG plate with variation of the material stiffness through the length for different  $E_c/E_m$  ratios ( $p=2$ )

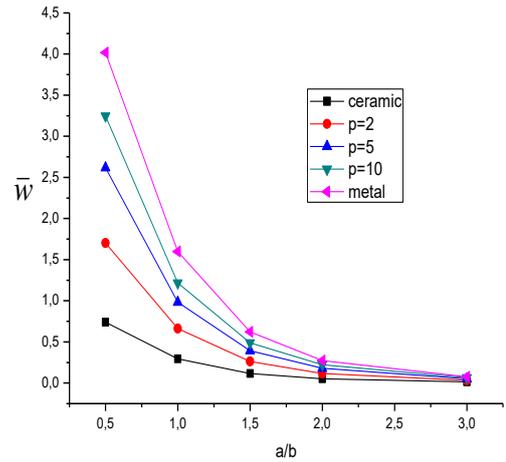


Fig. 4 Dimensionless center deflection  $\bar{w}$  as a function of the aspect ratio ( $a/b$ ) of an FGM plate ( $a/h=10$ )

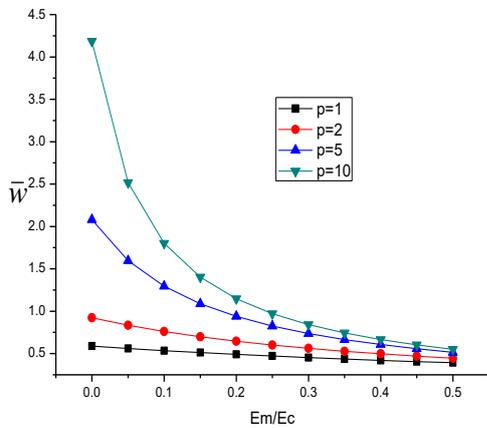


Fig. 3 The effect of material anisotropy on the dimensionless center deflection  $\bar{w}$  of an FGM plate for different values of  $p$

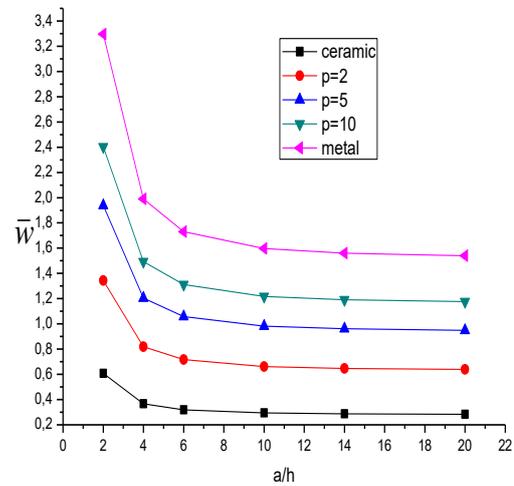


Fig. 5 Dimensionless center deflection  $\bar{w}$  as a function of the side to-thickness ratio ( $a/h$ ) of an FGM square plate

subjected to uniformly and sinusoidal distributed loads are presented.

Table 2 shows the calculated results of non-dimensional displacement and stress components of the Homogeneous square plate as compared with the published results of a generalized 2D shear deformation theory by Zenkour (2006) and Akavci (2015), where  $\epsilon_z=0$ . It can be seen from the table that the present 2D theory results are in excellent agreement with the 2D theory results of Zenkour (2006) and Akavci (2015). In addition, the present quasi-3D theory yields more accurate results than those obtained by the other two theories.

In Tables 3 and 4, the non-dimensional displacement and stresses of an  $Al/Al_2O_3$  FGM square plate subjected to sinusoidal distributed load are presented for different values of the power-law index. Table 3 shows the calculated results of non-dimensional displacement and stress components of the square FGM plate under uniform load as compared with the results of a generalized 2D shear deformation theory. It is well known that, the quasi-3D theory gives more accurate results than those obtained by the 2D theory where  $\epsilon_z=0$ . However, Table 3 demonstrates that these two theories give almost the same results. This is due to the used thickness

ratio ( $a/h=10$ ) which is not appropriate for very thick plates. This table also shows that, the transverse displacement  $w$  and the shear stresses  $\bar{\tau}_{xz}$  and  $\bar{\tau}_{xy}$  increase with the increasing value of power law index  $p$ .

Table 4 presents the non-dimensional in-plane stresses  $\bar{\sigma}_x$  and non-dimensional transverse displacements  $\bar{w}$  of a square plate for different  $a/h$  ratios. The present results are compared with the present 2D higher order shear deformation theory. The present quasi-3D that includes both transverse shear and normal deformations give accurate results than those given by the other theory.

The non-dimensional displacements and stresses are presented in Tables 5-6 for various values of aspect ratio  $b/a$ , thickness ratio  $a/h$  and exponent value  $p$ . Table 5 presents the central transverse displacements of the very thick EGM plates. The obtained results are compared with the 2D theory. Since the proposed theory includes the thickness-stretching effect, the results are accurate for thick plates. Meanwhile, 2D theory which do not include the thickness stretching effect overestimate the results. In Table 6, the calculated non-dimensional stresses are presented as

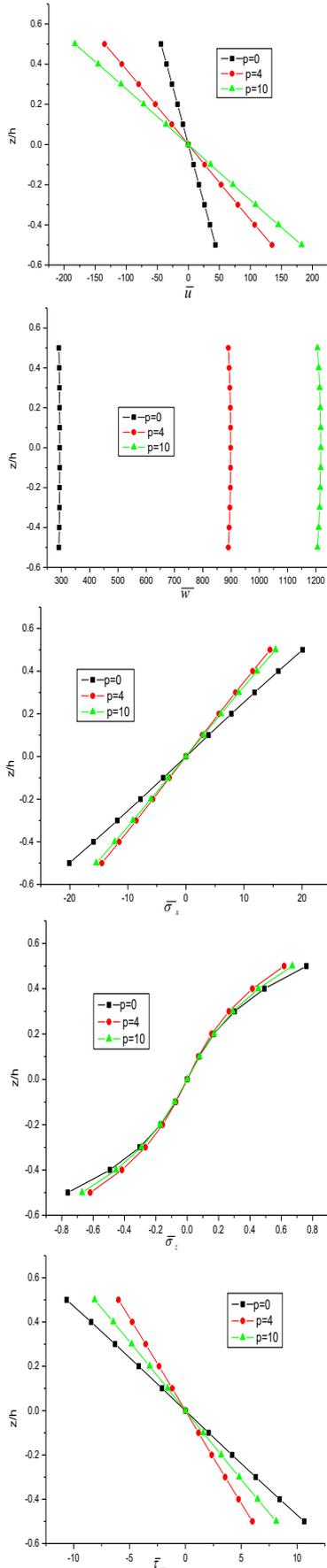


Fig. 6 The distributions of the non-dimensional displacement and stresses of FG plate ( $a/b=1, a/h=10$ )

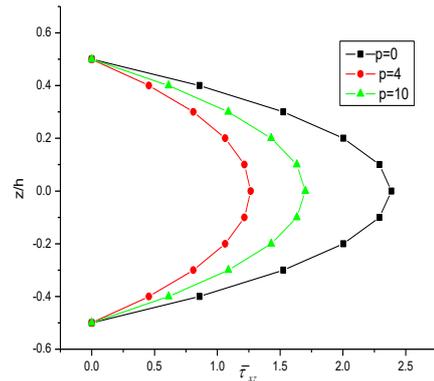


Table 6 Continued

compared with the 2D theory. It is evident from the tables that the present computations are in good agreement with the quasi-3D solutions.

Fig. 2 presents the deflection of the FG plate with variation of the material stiffness through the length for different  $E_c/E_m$  ratios. Generally, it is reported that an increase in the stiffness ratio,  $E_c/E_m$ , increase the deflection of the FG plate with property variation through the length, which shows the effect of different  $E_c/E_m$  ratios on the deflection of the FG plate.

On the whole stiffness of the FG plate is robustly associated with the direction of the material stiffness variation and the correct selection of the modulus ratio  $E_c/E_m$ . The modulus ratio also influences the location of maximum deflection. As the  $E_c/E_m$  ratio increases, the location of the maximum deflection moves towards the left side of the plate (dominated side). By using this aspect, the location of the maximum deflection, the stiffness of the FG plate can be controlled to meet the desired application with specific performance.

The center deflections of the simply supported FG square plate are compared in Fig. 3 for various ratios of moduli,  $E_m/E_c$  (for a given thickness,  $a/h=10$ ). In other words, the deflections are computed and compared for plates with different ceramic-metal mixtures. It is clear that the deflections decrease smoothly as the volume fraction exponent decreases, and decrease as the ratio of metal-to-ceramic moduli increases.

Figs. 4 and 5 show the variation of the center deflection for various power law exponent  $p$  and with different aspect and side-to-thickness ratios, respectively. The FG plate deflection is between those of plate made of ceramic ( $Al_2O_3$ ) and metal (Al). It can be observed that the deflection of metal rich plates is larger when compared to ceramic rich FGM plates, which can be attributed to the fact that the Young's modulus of ceramic ( $Al_2O_3$ , 380 GPa) is higher than that of metal (Al, 70 GPa). Hence for FG plates, the transverse deflection decreases as the power law exponent  $p$  decreases. In addition, the deflection of the FGM plate decreases as the aspect ratio increases, whereas it may be unchanged as the side-to-thickness ratio increases.

The stress and displacement distributions through the thickness of  $Al/Al_2O_3$  FGM square plate, under sinusoidal load, are presented in Fig. 6. The results are plotted for various values of power law index  $p$ . According to Fig. 6, it

is important to note that, through the thickness distributions of in-plane stresses  $\bar{\sigma}_x$  and  $\bar{\tau}_{xy}$  are linear for FG plate with in-plane variation of material stiffness along the length. The variation of the in-plane stress  $\bar{\sigma}_x$  at the mid-plane ( $xy$ ) of the FG plate with in-plane variation of material stiffness along the length is depicted in Fig. 6. The maximum tensile and compressive stresses occur at the bottom and top surfaces, respectively. It is observed that the stress  $\bar{\sigma}_x$  also becomes zero at the mid-surface of the thickness ( $z=0$ ) for FG plates with property variation through the length, which is the same trend for the case with constant property (homogenous plate).

## 5. Conclusions

In this work, the bending response of thick FG plates with power-law variation of volume fraction within the length is studied using a quasi-3D hyperbolic shear deformation theory. The exactitude of the present model has been confirmed for bending investigation of a simply supported FG plate under uniformly and sinusoidal distributed loads. From this study, it is observed that contrary to the case of a homogeneous plate, the maximum deflection does not found in the middle of the FG plate with in-plane distribution of material rigidity. It is also noticed that the overall rigidity of the FG plate strongly depends on the direction of the variation of rigidity of the material and the ratio between Young's moduli ( $E_c/E_m$ ) of the two phases. In addition, the location of the maximum transverse displacement does not depend on the thickness of the FG plate and the maximum deflection increases with an increase in the power index ( $p$ ). The effect of boundary conditions can be considered in future using the same methodology as is described by Ait Amar Meziane *et al.* (2014) and Abdelaziz *et al.* (2017).

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